The presence of a crack on a pressurized cylindrical shell structure leads to "complex stress and displacements fields resulting in nonlinear out-of-plane deformations" [1]. In other words, the single curved geometry and pressure differential causes a longitudinal crack to *bulge-out* or *protrude* from the original contour. This change in geometry of "bulging effect" significantly increases the stress-intensity factor (SIF) at the crack tips. The effects of this loading condition on composite laminae can trigger several types of failure mechanisms.

"One measure of the bulging effect is the **bulging factor**, which is the ratio of the SIF at the tip of a longitudinal crack in the fuselage to the SIF for the same crack in a flat panel. The damage tolerance design philosophy requires realistic stress state determination in the vicinity of cracks in airframe fuselages."[1]

Bulging factor emerges as a result of the out-of plane deformations of the surface of a crack on a pressurized fuselage structure. The representation of this phenomenon becomes rather complex due to the biaxial and internal pressure load and structural configuration [1] (Figure 1).



Figure 1 - (a) Comparison between a crack in plane stress condition and a crack on a curved shell; (b) mesh at the crack-tip for the Modified Crack Closure Integral (MCCI) technique; geometric parameters of the shell [3]

For the case of unstiffened shell structures, the bulging factor can be defined as the ratio of stress-intensity (SIF) of a curved shell to the stress-intensity factor of a flat panel:

$$\beta = \frac{K_{curved}}{K_{flat}}$$

References:

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